

Amendments to the Claims

Claims 1-8. (Canceled)

Claim 9. (New) A seal arrangement for a gas turbine, in particular for an aircraft engine compressor, for sealing a gap between a radially internally located end of a guide vane of a guide vane ring and a rotor, wherein the rotor includes at least two seal projections, positioned at an axial distance relative to each other, in a circumferential direction of the rotor, the seal projections providing a seal of the gap in combination with intake linings configured as honeycomb structures and associated with the radially internally located end of the guide vane, wherein the seal projections are inclined in an axial direction toward a side of higher pressure, and wherein, in a space limited by the two seal projections and the intake linings, at least one recirculation structure is provided and oriented toward the side of higher pressure.

Claim 10. (New) The seal arrangement according to Claim 9, wherein the recirculation structure is integrated in a radially internally located platform of the guide vane of the guide vane ring.

Claim 11. (New) The seal arrangement according to Claim 9, wherein the seal projections are configured as seal fins.

Claim 12. (New) The seal arrangement according to Claim 9, wherein a honeycomb of the honeycomb structures is configured such it is open in a direction toward the seal projections.

Claim 13. (New) The seal arrangement according to Claim 9, wherein the seal projections and intake linings have different radii, wherein an outer radii of the seal projections, as well as an inner radii of the intake linings, increase in the direction toward the side of higher pressure.

Claim 14. (New) A turbocompressor in axial construction and/or diagonal construction and/or radial construction, comprising a seal arrangement according to Claim 9.

Claim 15. (New) An aircraft engine comprising a turbocompressor according to Claim 14.

Claim 16. (New) A stationary gas turbine comprising a turbocompressor according to Claim 14.

Claim 17. (New) A seal for a gas turbine, comprising:
at least two seal projections disposed on a rotor;
at least two intake linings on a radially internal end of a stationary guide vane, wherein the at least two intake linings are configured as honeycomb structures and are disposed opposite the at least two seal projections; and
a recirculation structure disposed on the radially internal end of the stationary guide vane and between the at least two seal projections on the rotor.